



適航指令發布單

Airworthiness Directive Issuance Form

民航局AD編號 AD number	CAA-2024-10-018	發布日期 Date issued	2024/11/01												
適用之航空產品 Applied to (models, serial numbers or part numbers, as applicable)	Airbus A318-111, A318-112, A318-121, A318-122, A319-111, A319-112, A319-113, A319-114, A319-115, A319-131, A319-132, A319-133, A320-211, A320-212, A320-214, A320-215, A320-216, A320-231, A320-232, A320-233, A321-111, A321-112, A321-131, A321-211, A321-212, A321-213, A321-231 and A321-232 aeroplanes, all manufacturer serial numbers, except: - A319 aeroplanes on which Airbus modification (mod) 28238, mod 28162 and mod 28342 have been embodied in production; and - A318 aeroplanes on which mod 39195 has been embodied in production, or Airbus Service Bulletin (SB) A320-00-1219 has been embodied in service.														
主旨摘要 Subject	Fuselage - Inner Cap and Frame Flange at Frame 68 Stringer 22 - Inspections														
民航局 CAA <input type="radio"/> 本國產品 Native product <input type="radio"/> 其他個案 Other	設計國民航主管機構 Original Authority <table><tr><td><input type="radio"/> FAA</td><td><input type="radio"/> Germany LBA</td></tr><tr><td><input checked="" type="radio"/> EASA</td><td><input type="radio"/> CAA-NL</td></tr><tr><td><input type="radio"/> Brazil</td><td><input type="radio"/> UK CAA</td></tr><tr><td><input type="radio"/> Transport Canada Civil Aviation</td><td><input type="radio"/> Japan CAB</td></tr><tr><td><input type="radio"/> DGAC</td><td><input type="radio"/> CAA of Israel</td></tr><tr><td></td><td><input type="radio"/> Other_____</td></tr></table>			<input type="radio"/> FAA	<input type="radio"/> Germany LBA	<input checked="" type="radio"/> EASA	<input type="radio"/> CAA-NL	<input type="radio"/> Brazil	<input type="radio"/> UK CAA	<input type="radio"/> Transport Canada Civil Aviation	<input type="radio"/> Japan CAB	<input type="radio"/> DGAC	<input type="radio"/> CAA of Israel		<input type="radio"/> Other_____
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<input type="radio"/> Transport Canada Civil Aviation	<input type="radio"/> Japan CAB														
<input type="radio"/> DGAC	<input type="radio"/> CAA of Israel														
	<input type="radio"/> Other_____														
	設計國AD編號 Original AD number	2024-0210													
	<p>1. 直接採用原AD之內容? (Is the original AD directly adopted?)</p> <p><input checked="" type="radio"/> 是(Yes) <input type="radio"/> 否(No)_</p> <p>a. 生效日期另訂為(Re-specify the effective date as):</p> <p>b. 執行時限另訂為(Re-specify the compliance time or period as):</p> <p>2. 使用人是否需要將AD執行結果向民航局提出報告? (Do users need to report the status of compliance to the CAA?)</p> <p><input type="radio"/> 需要(Yes) <input checked="" type="radio"/> 不需要(No)</p>														
備註 Note	This AD supersedes EASA AD 2022-0030(CAA-2022-03-001) dated 25 February 2022.														

註： 1. AD內容後附。
2. 航空器產品使用人得向民航局提出豁免、替代符合方法、執行時限之展延之申請。
3. 如有任何問題，請聯絡交通部民用航空局初始適航科。Tel：(02)2349-6330 / 6332, Fax：(02)2545-8464,
adcaa@mail.caa.gov.tw

Note：1. The AD text is enclosed.
2. Exemption, an alternative method of compliance or adjustment of the compliance time may be proposed to the CAA for approval.
3. For further information, please contact Civil Aviation Administration on Tel：(02)2349-6330 / 6332, Fax：(02)2545-8464, adcaa@mail.caa.gov.tw



Airworthiness Directive

AD No.: 2024-0210

Issued: 29 October 2024

Note: This Airworthiness Directive (AD) is issued by EASA, acting in accordance with Regulation (EU) 2018/1139 on behalf of the European Union, its Member States and of the European third countries that participate in the activities of EASA under Article 129 of that Regulation.

This AD is issued in accordance with Regulation (EU) 748/2012, Part 21.A.3B. In accordance with Regulation (EU) 1321/2014 Annex I Part M.A.301, or Annex Vb Part ML.A.301, as applicable, the continuing airworthiness of an aircraft shall be ensured by accomplishing any applicable ADs. Consequently, no person may operate an aircraft to which an AD applies, except in accordance with the requirements of that AD, unless otherwise specified by the Agency [Regulation (EU) 1321/2014 Annex I Part M.A.303, or Annex Vb Part ML.A.303, as applicable] or agreed with the Authority of the State of Registry [Regulation (EU) 2018/1139, Article 71 exemption].

Design Approval Holder's Name:

AIRBUS S.A.S.

Type/Model designation(s):

A318, A319, A320 and A321 aeroplanes

Effective Date: 12 November 2024

TCDS Number(s): EASA.A.064

Foreign AD: Not applicable

Supersedure: This AD supersedes EASA AD 2022-0030 dated 25 February 2022.

ATA 53 – Fuselage – Inner Cap and Frame Flange at Frame 68 Stringer 22 – Inspections

Manufacturer(s):

Airbus, formerly Airbus Industrie

Applicability:

Airbus A318-111, A318-112, A318-121, A318-122, A319-111, A319-112, A319-113, A319-114, A319-115, A319-131, A319-132, A319-133, A320-211, A320-212, A320-214, A320-215, A320-216, A320-231, A320-232, A320-233, A321-111, A321-112, A321-131, A321-211, A321-212, A321-213, A321-231 and A321-232 aeroplanes, all manufacturer serial numbers, except:

- A319 aeroplanes on which Airbus modification (mod) 28238, mod 28162 and mod 28342 have been embodied in production; and
- A318 aeroplanes on which mod 39195 has been embodied in production, or Airbus Service Bulletin (SB) A320-00-1219 has been embodied in service.

Definitions:

For the purpose of this AD, the following definitions apply:

Inspection area: Inner cap and web horizontal flange at frame 68 (left-hand (LH) and right-hand (RH) sides) at level of stringer 22 and around the door stop 1 nuts, aft passenger/crew door, LH and RH sides.



The SB: Airbus SB A320-53-1491 Revision 02.

Groups: Group 1 aeroplanes are those on which Airbus repair part having Part Number (P/N) R534-20799 is installed or any other repair part approved by Airbus is embodied on the inspection area.

Group 2 aeroplanes are those which are not Group 1.

An aeroplane may belong to both Groups, one for each applicable configuration of the inspection area (RH/LH side).

Reason:

Cracks have been reported in the inner cap and web horizontal flange at frame 68 (LH and RH sides) at level of stringer 22 during accomplishment of inspections required by EASA AD 2016-0238 (later superseded by EASA AD 2021-0242).

This condition, if not detected and corrected, could reduce the structural integrity of the fuselage.

To address this potential unsafe condition, Airbus issued the SB A320-53-1491 (later revised), and EASA issued AD 2022-0030 to require repetitive inspections of that area.

After that AD was issued, cracks have been found at the door stop fitting number 1 holes at frame 68, after the door stop fitting disassembly during accomplishment of inspections in accordance with the instructions of SB A320-53-1491 Revision 01.

Therefore, Airbus issued the SB, as defined in this AD, to include an additional inspection of the frame 68 door stop fitting number 1 holes. The inspection area is extended with an additional high-frequency eddy-current (HFEC) inspection to be performed on the FR68 around the door stop fitting number 1 nuts. The compliance times have been reassessed and the SB has been updated accordingly.

For the reasons described above, this AD retains the requirements of EASA AD 2022-0030, which is superseded, and requires repetitive special detailed inspection (SDI) of the inspection area.

Required Action(s) and Compliance Time(s):

Required as indicated by this AD, unless the action(s) required by this AD have been already accomplished:

Inspection(s):

- (1) Before exceeding the compliance time(s) as defined in Table 1 of this AD, and, thereafter, at intervals not to exceed the value as defined in Table 1 of this AD, as applicable, accomplish the SDI of each inspection area in accordance with the instructions of the SB.
- (2) If, before the effective date of this AD, any Airbus approved repair instructions have been issued, supplementing the instructions of SB A320-53-1491 at any revision for an inspection area, accomplish those instructions on that inspection area within the compliance time specified therein. Accomplishment of those inspections on an inspection area does not supersede the requirement of paragraph (1) of this AD for that inspection area.



- (3) If, before the effective date of this AD, any Airbus approved repair instructions have been issued, superseding the instructions of SB A320-53-1491 at any revision for an inspection area, accomplish those instructions on that inspection area within the compliance time specified therein. Accomplishment of those inspections on an inspection area supersedes the requirement of paragraph (1) of this AD for that inspection area.

Table 1 – Initial SDI and Interval

Group	Inspection Thresholds (whichever occurs later)	Intervals
Group 1	Within 20 000 flight cycles (FC) after aeroplane first flight, or within 19 700 FC after the last inspection in accordance with the original issue or Revision 01 of the instructions of Airbus SB A320-53-1491, as applicable.	19 700 FC
Group 2	Within 20 000 FC after aeroplane first flight, or for aeroplanes without Airbus SB A320-53-1290 embodied at any revision: Within 16 200 FC after the last inspection in accordance with, whichever occurs first: - Airbus SB A320-53-1491 without findings, AND - Airbus SB A320-53-1288, or Airworthiness Limitation Item (ALI) task 534130 accomplished before 22 November 2021 [the effective date of EASA AD 2021-0242] without findings at FR68 Door stop 1, or, Airbus SB A320-53-1288 with findings and repaired in accordance with SRM task 53-41-12-300-009 at FR68 Door Stop 1, or for aeroplanes having SB A320-53-1290 Revision 03 or later Revision embodied: Within 16 200 FC after embodiment of SB A320-53-1290 Revision 03 or later Revision AND after last inspection in accordance with the Airbus SB A320-53-1491 without findings, whichever occurs first as applicable, or within 12 months after the effective date of this AD.	16 200 FC

Corrective Action(s):

- (4) If, during any inspection as required by paragraph (1), (2) or (3) of this AD, as applicable, discrepancies and/or cracks are detected, before next flight, accomplish the applicable corrective actions in accordance with the instructions of the SB, or contact Airbus for approved corrective action(s) instructions and, within the compliance time specified therein, accomplish those instructions accordingly, as applicable (see Note 1 of this AD).

Note 1: After embodiment of any Airbus approved repair part on a Group 2 aeroplane, that aeroplane is considered a Group 1 aeroplane for that repaired inspection area.



Credit:

- (5) For Group 1 aeroplanes: SDI(s) accomplished on an inspection area, before the effective date of this AD, in accordance with the instructions of the Airbus SB A320-53-1491 at original issue or Revision 01, as applicable, is acceptable to comply with the requirements of paragraph (1) of this AD for that inspection area.

Note 2: Credit provided by paragraph (5) can only be taken for inspection areas, where instructions in the initial issue or Revision 01 of the SB exist.

Terminating Action:

- (6) Accomplishment of corrective action(s) on an inspection area as required by paragraph (4) of this AD does not constitute terminating action for the repetitive SDI as required by paragraph (1), (2) or (3) of this AD for that inspection area, unless specified otherwise in the instructions provided by Airbus.

Reporting:

- (7) Within 90 days after accomplishment of each DET as required by paragraph (1) of this AD, or after the effective date of this AD, whichever occurs later, report the findings to Airbus. Using the inspection report attached to the SB is an acceptable method to comply with this requirement.

Ref. Publications:

Airbus SB A320-53-1491 original issue dated 14 August 2020, or Revision 01 dated 2 May 2022, or Revision 02 dated 30 July 2024.

The use of later approved revisions of the above-mentioned document is acceptable for compliance with the requirements of this AD.

Remarks:

1. If requested and appropriately substantiated, EASA can approve Alternative Methods of Compliance for this AD.
2. This AD was posted on 30 September as PAD 24-113 for consultation until 14 October 2024. The Comment Response Document can be found in the [EASA Safety Publications Tool](#), in the compressed (zipped) file attached to the record for this AD.
3. Enquiries regarding this AD should be referred to the EASA Safety Information Section, Certification Directorate. E-mail: ADs@easa.europa.eu.
4. Information about any failures, malfunctions, defects or other occurrences, which may be similar to the unsafe condition addressed by this AD, and which may occur, or have occurred on a product, part or appliance not affected by this AD, can be reported to the [EU aviation safety reporting system](#). This may include reporting on the same or similar components, other than those covered by the design to which this AD applies, if the same unsafe condition can exist or may develop on an aircraft with those components installed. Such components may be installed under an FAA Parts Manufacturer Approval (PMA), Supplemental Type Certificate (STC) or other modification.



5. For any question concerning the technical content of the requirements in this AD, please contact: AIRBUS – Airworthiness Office – 1IASA; E-mail: account.airworth-eas@airbus.com.

