



# 適航指令發布單 Airworthiness Directive Issuance Form

民航局AD編號 AD number	CAA-2024-01-006	發布日期 Date issued	2024/01/17												
適用之航空產品 Applied to (models, serial numbers or part numbers, as applicable)	Airbus A330-201, A330-202, A330-203, A330-223, A330-223F, A330-243, A330-243F, A330-301, A330-302, A330-303, A330-321, A330-322, A330-323, A330-341, A330-342, A330-343, and A330-941 aeroplanes, all manufacturer serial numbers (MSN) up to MSN 1919 inclusive, except MSN 1915; and Airbus A340-211, A340-212, A340-213, A340-311, A340-312 and A340-313 aeroplanes, all MSN.														
主旨摘要 Subject	Flight Controls - Trimmable Horizontal Stabilizer Actuator / Electric Load Sensing Device - Modification														
民航局 CAA  <input type="radio"/> 本國產品 Native product  <input type="radio"/> 其他個案 Other	設計國民航主管機構 Original Authority <table><tr><td><input type="radio"/> FAA</td><td><input type="radio"/> Germany LBA</td></tr><tr><td><input checked="" type="radio"/> EASA</td><td><input type="radio"/> CAA-NL</td></tr><tr><td><input type="radio"/> Brazil</td><td><input type="radio"/> UK CAA</td></tr><tr><td><input type="radio"/> Transport Canada Civil Aviation</td><td><input type="radio"/> Japan CAB</td></tr><tr><td><input type="radio"/> DGAC</td><td><input type="radio"/> CAA of Israel</td></tr><tr><td></td><td><input type="radio"/> Other_____</td></tr></table>			<input type="radio"/> FAA	<input type="radio"/> Germany LBA	<input checked="" type="radio"/> EASA	<input type="radio"/> CAA-NL	<input type="radio"/> Brazil	<input type="radio"/> UK CAA	<input type="radio"/> Transport Canada Civil Aviation	<input type="radio"/> Japan CAB	<input type="radio"/> DGAC	<input type="radio"/> CAA of Israel		<input type="radio"/> Other_____
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<input type="radio"/> Brazil	<input type="radio"/> UK CAA														
<input type="radio"/> Transport Canada Civil Aviation	<input type="radio"/> Japan CAB														
<input type="radio"/> DGAC	<input type="radio"/> CAA of Israel														
	<input type="radio"/> Other_____														
	設計國AD編號 Original AD number	2024-0016													
	1. 直接採用原AD之內容? (Is the original AD directly adopted?) <input checked="" type="radio"/> 是(Yes) <input type="radio"/> 否(No)_ a. 生效日期另訂為(Re-specify the effective date as): b. 執行時限另訂為(Re-specify the compliance time or period as):  2. 使用人是否需要將AD執行結果向民航局提出報告? (Do users need to report the status of compliance to the CAA?) <input type="radio"/> 需要(Yes) <input checked="" type="radio"/> 不需要(No)														
備註 Note	This AD supersedes EASA AD 2022-0039 (CAA-2022-03-009) dated 08 March 2022.														

註： 1. AD內容後附。

2. 航空器產品使用人得向民航局提出豁免、替代符合方法、執行時限之展延之申請。

3. 如有任何問題，請聯絡交通部民用航空局初始適航科。Tel：(02)2349-6330 / 6332, Fax：(02)2545-8464,

[adcaa@mail.caa.gov.tw](mailto:adcaa@mail.caa.gov.tw)

Note： 1. The AD text is enclosed.

2. Exemption, an alternative method of compliance or adjustment of the compliance time may be proposed to the CAA for approval.

3. For further information, please contact Civil Aeronautics Administration on Tel：(02)2349-6330 / 6332,

Fax：(02)2545-8464, [adcaa@mail.caa.gov.tw](mailto:adcaa@mail.caa.gov.tw)



## Airworthiness Directive

**AD No.:** 2024-0016

**Issued:** 11 January 2024

Note: This Airworthiness Directive (AD) is issued by EASA, acting in accordance with Regulation (EU) 2018/1139 on behalf of the European Union, its Member States and of the European third countries that participate in the activities of EASA under Article 129 of that Regulation.

This AD is issued in accordance with Regulation (EU) 748/2012, Part 21.A.3B. In accordance with Regulation (EU) 1321/2014 Annex I Part M.A.301, or Annex Vb Part ML.A.301, as applicable, the continuing airworthiness of an aircraft shall be ensured by accomplishing any applicable ADs. Consequently, no person may operate an aircraft to which an AD applies, except in accordance with the requirements of that AD, unless otherwise specified by the Agency [Regulation (EU) 1321/2014 Annex I Part M.A.303, or Annex Vb Part ML.A.303, as applicable] or agreed with the Authority of the State of Registry [Regulation (EU) 2018/1139, Article 71 exemption].

**Design Approval Holder's Name:**

AIRBUS S.A.S.

**Type/Model designation(s):**

A330 and A340 aeroplanes

**Effective Date:** 25 January 2024

**TCDS Number(s):** EASA.A.004, EASA.A.015

**Foreign AD:** Not applicable

**Supersedure:** This AD supersedes EASA AD 2022-0039 dated 08 March 2022.

### ATA 27 – Flight Controls – Trimmable Horizontal Stabilizer Actuator / Electric Load Sensing Device – Modification

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**Manufacturer(s):**

Airbus, formerly Airbus Industrie

**Applicability:**

Airbus A330-201, A330-202, A330-203, A330-223, A330-223F, A330-243, A330-243F, A330-301, A330-302, A330-303, A330-321, A330-322, A330-323, A330-341, A330-342, A330-343, and A330-941 aeroplanes, all manufacturer serial numbers (MSN) up to MSN 1919 inclusive, except MSN 1915; and

Airbus A340-211, A340-212, A340-213, A340-311, A340-312 and A340-313 aeroplanes, all MSN.

**Definitions:**

For the purpose of this AD, the following definitions apply:

**The SB1a:** Airbus Service Bulletin (SB) A330-27-3237 and SB A340-27-4213, as applicable, both at original issue, and the Airbus Repair and Design Approval Form (RDAF) 80874366/013/2021#A for A330 aeroplanes or RDAF 80874366/022/2021#A for A340 aeroplanes, as applicable (for aeroplanes to which the RDAF applies).



**The SB1b:** Airbus Service Bulletin (SB) A330-27-3237 and SB A340-27-4213, as applicable, both at Revision 01 or later.

**The SB2:** Airbus SB A330-27-3234 and SB A340-27-4214, as applicable.

**Groups:** Group 1 aeroplanes are all MSN up to MSN 1789 inclusive.  
Group 2 aeroplanes are MSN 1790 to 1919 inclusive.

#### Reason:

The upper and lower attachments of the Trimmable Horizontal Stabilizer Actuator (THSA) have a primary load path (PLP) and a secondary load path (SLP), the latter of which is only engaged in case of PLP failure. When the SLP is engaged, the THSA should stall, and an indication should be provided to the flight crew, activated by position monitoring. It has been demonstrated by recent tests that, when the upper SLP is engaged, the unit might not stall, consequently with no indication of SLP engagement.

This condition, if not corrected, could lead to damage on the upper THSA SLP attachment, with consequent mechanical disconnection of the THSA, possibly resulting in loss of control of the aeroplane.

To initially address this potential unsafe condition, Airbus developed a method to inspect the upper THSA attachments parts and the PLP and SLP fuselage attachment points, and EASA issued AD 2017-0044 to require those repetitive inspections and, depending on findings, accomplishment of applicable corrective action(s). That AD was later cancelled, as the requirements were transferred into the applicable Airworthiness Limitation Sections (ALS) for the affected type designs, for which EASA published AD 2019-0047 and AD 2019-0048.

After those ADs were issued, Airbus designed an Electric Load Sensing Device (ELSD), to detect the engagement on the SLP, even in absence of a THSA stall. Consequently, Airbus published the SB1a, providing instructions for installation of the ELSD wiring provisions, and the SB2, providing instructions for ELSD installation and activation, and EASA issued AD 2022-0039 to require to modify the THSA installation, implementing ELSD wiring provision and installing and activating the ELSD.

Since that AD was issued, it has been determined that the SB1a cannot be accomplished on certain aeroplanes, and Airbus initially issued several adaptations to provide additional instructions and corrections, and eventually the SB1b.

For the reasons described above, this AD retains the requirements of EASA AD 2022-0039, which is superseded, but refers to the SB1b, and requires additional work for certain aeroplanes.

#### Required Action(s) and Compliance Time(s):

Required as indicated by this AD, unless the actions required by this AD have been already accomplished:



**Modification(s):**

Within 48 months after 22 March 2022 [the effective date of EASA AD 2022-0039], accomplish the following:

- (1) For Group 1 aeroplanes: Install the wiring for the ELSD in accordance with the instructions of the SB1b.
- (2) For Group 1 and Group 2 aeroplanes: Install and activate the ELSD in accordance with the instructions of the SB2.

**Additional Work:**

- (3) For Group 1 aeroplanes that, before the effective date of this AD, have been modified in accordance with the instructions of the SB1a: Within 48 months after 22 March 2022 [the effective date of EASA AD 2022-0039], accomplish the additional work as identified in, and in accordance with the instruction of, the SB1b, as applicable.

**Ref. Publications:**

Airbus SB A330-27-3234 original issue dated 05 March 2019 and Revision 01 dated 12 October 2020.

Airbus SB A330-27-3237 original issue dated 12 October 2020 and Revision 01 dated 20 June 2023.

Airbus SB A340-27-4213 original issue dated 12 October 2020 and Revision 01 dated 20 June 2023.

Airbus SB A340-27-4214 original issue dated 12 October 2020.

Airbus RDAF 80874366/013/2021#A dated 12 February 2021.

Airbus RDAF 80874366/022/2021#A dated 19 April 2021.

The use of later approved revisions of the above-mentioned documents is acceptable for compliance with the requirements of this AD.

**Remarks:**

1. If requested and appropriately substantiated, EASA can approve Alternative Methods of Compliance for this AD.
2. This AD was posted on 12 December 2023 as PAD 23-141 for consultation until 09 January 2024. No comments were received during the consultation period.
3. Enquiries regarding this AD should be referred to the EASA Safety Information Section, Certification Directorate. E-mail: [ADs@easa.europa.eu](mailto:ADs@easa.europa.eu).
4. Information about any failures, malfunctions, defects or other occurrences, which may be similar to the unsafe condition addressed by this AD, and which may occur, or have occurred on a product, part or appliance not affected by this AD, can be reported to the [EU aviation safety reporting system](#). This may include reporting on the same or similar components, other than



those covered by the design to which this AD applies, if the same unsafe condition can exist or may develop on an aircraft with those components installed. Such components may be installed under an FAA Parts Manufacturer Approval (PMA), Supplemental Type Certificate (STC) or other modification.

5. For any question concerning the technical content of the requirements in this AD, please contact: AIRBUS – 1IAL (Airworthiness Office), E-mail: [airworthiness.A330-A340@airbus.com](mailto:airworthiness.A330-A340@airbus.com).

