



適航指令發布單
Airworthiness Directive Issuance Form

民航局 AD 編號 AD number	CAA-2021-06-006	發布日期 Date issued	2021/6/21												
適用之航空產品 Applied to (models, serial numbers or part numbers, as applicable)	Airbus A350-941 and A350-1041 aeroplanes, all manufacturer serial numbers, except those on which Airbus modification (mod) 112721, mod 112722, or mod 113501 has been embodied in production.														
主旨摘要	Fuselage - Trimmable Horizontal Stabilizer Fittings and Bearing Assembly Attachment Interface - Inspection														
民航局 CAA <input type="checkbox"/> 本國產品 Native products <input type="checkbox"/> 其他個案 Other	<div style="text-align: center;">設計國民航主管機構 Original Authorities</div> <table style="width: 100%;"><tr><td><input type="checkbox"/> FAA</td><td><input type="checkbox"/> Germany LBA</td></tr><tr><td><input checked="" type="checkbox"/> EASA</td><td><input type="checkbox"/> CAA-NL</td></tr><tr><td><input type="checkbox"/> Brazil</td><td><input type="checkbox"/> UK CAA</td></tr><tr><td><input type="checkbox"/> Transport Canada Civil Aviation</td><td><input type="checkbox"/> Japan CAB</td></tr><tr><td><input type="checkbox"/> DGAC</td><td><input type="checkbox"/> CAA of Israel</td></tr><tr><td colspan="2"><input type="checkbox"/> Other _____</td></tr></table>			<input type="checkbox"/> FAA	<input type="checkbox"/> Germany LBA	<input checked="" type="checkbox"/> EASA	<input type="checkbox"/> CAA-NL	<input type="checkbox"/> Brazil	<input type="checkbox"/> UK CAA	<input type="checkbox"/> Transport Canada Civil Aviation	<input type="checkbox"/> Japan CAB	<input type="checkbox"/> DGAC	<input type="checkbox"/> CAA of Israel	<input type="checkbox"/> Other _____	
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<input type="checkbox"/> Other _____															
	設計國 AD 編號 Original AD number	2021-0141													
	1. 直接採用原 AD 之內容?(Is the original AD directly adopted?) <input checked="" type="checkbox"/> 是(Yes) <input type="checkbox"/> 否(No) _ a. 生效日期另訂為(Re-specify the effective date as) : _____ b. 執行時限另訂為(Re-specify the compliance time or period as) : _____ <input type="checkbox"/> 2. 使用人是否需要將 AD 執行結果向民航局提出報告?(Do <input type="checkbox"/> Users need to report the status of compliance to the CAA?) <input type="checkbox"/> 是(Yes) <input checked="" type="checkbox"/> 否(No)														
備註 Note	This AD supersedes EASA AD 2019-0206(CAA-2019-08-006) dated 20 August 2019.														

註： 1. AD 內容後附。
2. 航空器產品使用人得向民航局提出豁免、替代符合方法、執行時限之展延之申請。
3. 如有任何問題，請聯絡交通部民用航空局初始適航科。Tel：(02)2349-6331~3, Fax：(02)2545-8464, e-mail：adcaa@mail.caa.gov.tw

Note： 1. The AD text is enclosed.
2. Exemption, an alternative method of compliance or adjustment of the compliance time may be proposed to the CAA for approval.
3. For further information, please contact Civil Aeronautics Administration on Tel：(02)2349-6331~3, Fax：(02)2545-8464, e-mail：adcaa@mail.caa.gov.tw



Airworthiness Directive

AD No.: 2021-0141

Issued: 15 June 2021

Note: This Airworthiness Directive (AD) is issued by EASA, acting in accordance with Regulation (EU) 2018/1139 on behalf of the European Union, its Member States and of the European third countries that participate in the activities of EASA under Article 129 of that Regulation.

This AD is issued in accordance with Regulation (EU) 748/2012, Part 21.A.3B. In accordance with Regulation (EU) 1321/2014 Annex I, Part M.A.301, the continuing airworthiness of an aircraft shall be ensured by accomplishing any applicable ADs. Consequently, no person may operate an aircraft to which an AD applies, except in accordance with the requirements of that AD, unless otherwise specified by the Agency [Regulation (EU) 1321/2014 Annex I, Part M.A.303] or agreed with the Authority of the State of Registry [Regulation (EU) 2018/1139, Article 71 exemption].

Design Approval Holder's Name:

AIRBUS

Type/Model designation(s):

A350 aeroplanes

Effective Date: 29 June 2021

TCDS Number(s): EASA.A.151

Foreign AD: Not applicable

Supersedure: This AD supersedes EASA AD 2019-0206 dated 20 August 2019.

ATA 53 – Fuselage – Trimmable Horizontal Stabilizer Fittings and Bearing Assembly Attachment Interface – Inspection

Manufacturer(s):

Airbus

Applicability:

Airbus A350-941 and A350-1041 aeroplanes, all manufacturer serial numbers, except those on which Airbus modification (mod) 112721, mod 112722, or mod 113501 has been embodied in production.

Definitions:

For the purpose of this AD, the following definitions apply:

Aeroplane date of manufacture: The date of transfer of title (ownership) at the time of first delivery to an operator, which is referenced in Airbus documentation.

The inspection SB: Airbus Service Bulletin (SB) A350-53-P032 Revision 01 (including additional information provided by Airbus SB Information Transmission (SBIT) 19-0014), or Revision 02.

The modification SB: Airbus SB A350-53-P037.



Groups: Group 1 aeroplanes are those on which Airbus mod 113102 has not been embodied in production. Group 2 aeroplanes are those on which Airbus mod 113102 has been embodied in production.

Reason:

Occurrences were reported of some aeroplanes where, during in-service lubrication of the trimmable horizontal stabilizer (THS) upper and lower frame attachment bearings at fuselage frame (FR) 102, evidence was found of rotation of an attachment fitting bearing assembly, left-hand (LH) and/or right-hand (RH) side. Rotation itself is not detrimental in terms of structural strength capabilities or THS functionality. However, rotation of the bearing assembly causes damage to the sealant bead, in which case water ingress may occur, thereby creating the risk of corrosion in the aluminium corner fitting.

This condition, if not detected and corrected, could lead to detachment of the THS from the fuselage, possibly resulting in loss of control of the aeroplane and injury to persons on the ground.

To address this unsafe condition, Airbus issued SB A350-53-P032 (original issue), providing inspection instructions to ensure that any visible damage and corrosion in the area around the THS lower corner fittings, bushing and bearing interface at FR102, LH and RH sides, are detected in time and repaired appropriately. Consequently, EASA issued AD 2018-0037 to require repetitive detailed inspections (DET) and, depending on findings, accomplishment of applicable corrective action(s).

After that AD was issued, and prompted by operator feedback, Airbus issued the inspection SB (Revision 01), as defined in this AD, to include instructions and tooling references to avoid removing section 19.1. Airbus also developed production mod 112721, 112722, 113102 and 113501 to remove the existing sealant beads previously applied to the bearing body and nut perimeter and to apply a new protection scheme with corrosion preventive compound and grease. Airbus mod 113501 was made available for in-service aeroplanes through the modification SB. In addition, the Effectivity of the inspection SB was expanded to include A350-1041 aeroplanes. Following issuance of Revision 01 of the inspection SB, as defined in this AD, Airbus published SBIT 19-0014 to confirm that no concurrent requirements exist, and Airbus issued Revision 02 of the inspection SB to include the information provided by SBIT 19-0014. Consequently, EASA issued AD 2019-0206, retaining the requirements of EASA AD 2018-0037, which was superseded, amending the repetitive inspections, adding pre-mod A350-1041 aeroplanes to the Applicability, excluding post-mod aeroplanes from the Applicability and introducing an optional terminating action given by the modification SB, as defined in this AD.

Since that AD was issued, it was identified that Airbus production mod 113102 had not been correctly applied. For retrofit purposes, Airbus issued SB A350-53-P067 (for Group 2 aeroplanes) to provide instructions for a one-time DET, and additional work instructions to ensure correct embodiment of Airbus mod 113102.

For the reasons described above, this AD retains the requirements of EASA AD 2019-0206, which is superseded, and adds post-mod 113102 aeroplanes to the Applicability to require a one-time DET and, depending on findings, accomplishment of associated corrective action(s).



Required Action(s) and Compliance Time(s):

Required as indicated, unless accomplished previously:

Re-statement of the Requirements of EASA AD 2019-0206:**Repetitive Inspections:**

- (1) For Group 1 aeroplanes: Before exceeding 36 months since the aeroplane date of manufacture, or within 3 months after 21 February 2018 [the effective date of EASA AD 2018-0037], whichever occurs later, and, thereafter, at intervals not to exceed 36 months, accomplish a DET of the fillet sealant in the lower and upper corner fittings and bearing assembly attachment interface at FR102, LH and RH sides, in accordance with the instructions of the inspection SB.
- (2) If, during any DET as required by paragraph (1) of this AD, any damage is found, before next flight, accomplish a DET of accessible area all around at lower and upper corner fittings, bushing and bearing interface at FR102, LH and/or RH sides, as applicable, in accordance with the instructions of the inspection SB.

Corrective Action(s):

- (3) If, during any DET as required by paragraph (2) of this AD, corrosion is found, before next flight, contact Airbus for approved repair instructions and, within the compliance time(s) specified in those instructions, accomplish those instructions accordingly.

Credit for Previous Action(s):

- (4) For Group 1 aeroplanes: Inspections and corrective actions on an aeroplane, accomplished before the effective date of this AD in accordance with the instructions of Airbus SB A350-53-P032 at original issue, are acceptable to comply with the initial DET requirements of this AD for that aeroplane.

Terminating Action:

- (5) Repair of an aeroplane as required by paragraph (3) of this AD does not constitute terminating action for the repetitive DET as required by paragraphs (1) or (2) of this AD for that aeroplane, unless specified otherwise in the approved repair instructions provided by Airbus.
- (6) For Group 1 aeroplanes: Modification of an aeroplane in accordance with the instructions of the modification SB constitutes terminating action for the repetitive DET as required by paragraph (1) of this AD for that aeroplane.

New Requirements of this AD:**Inspection:**

- (7) For Group 2 aeroplanes: Before exceeding 36 months since the aeroplane date of manufacture, or within 3 months after the effective date of this AD, whichever occurs later, accomplish a DET of the fillet sealant in the THS lower and upper corner fittings and bearing assembly attachment interface at fuselage FR102, LH and RH sides, in accordance with the instructions of Airbus SB A350-53-P067.



Corrective Action(s):

- (8) If, during the DET as required by paragraph (7) of this AD, any discrepancy is identified as described in Airbus SB A350-53-P067, before next flight, accomplish the associated corrective action(s) in accordance with the instructions of Airbus SB A350-53-P067.

Ref. Publications:

Airbus SB A350-53-P032 original issue dated 22 December 2017, or Revision 01 dated 12 November 2018, or Revision 02 dated 03 September 2020.

Airbus SB A350-53-P037 original issue dated 12 November 2018.

Airbus SB A350-53-P067 original issue dated 09 April 2021.

The use of later approved revisions of the above-mentioned documents is acceptable for compliance with the requirements of this AD.

Remarks:

1. If requested and appropriately substantiated, EASA can approve Alternative Methods of Compliance for this AD.
2. This AD was posted on 29 April 2021 as PAD 21-060 for consultation until 27 May 2021. The Comment Response Document can be found in the [EASA Safety Publications Tool](#), in the compressed (zipped) file attached to the record for this AD.
3. Enquiries regarding this AD should be referred to the EASA Safety Information Section, Certification Directorate. E-mail: ADs@easa.europa.eu.
4. Information about any failures, malfunctions, defects or other occurrences, which may be similar to the unsafe condition addressed by this AD, and which may occur, or have occurred on a product, part or appliance not affected by this AD, can be reported to the [EU aviation safety reporting system](#). This may include reporting on the same or similar components, other than those covered by the design to which this AD applies, if the same unsafe condition can exist or may develop on an aircraft with those components installed. Such components may be installed under an FAA Parts Manufacturer Approval (PMA), Supplemental Type Certificate (STC) or other modification.
5. For any question concerning the technical content of the requirements in this AD, please contact: AIRBUS A350 XWB, E-mail: continued-airworthiness.a350@airbus.com.

