



適航指令發布單

Airworthiness Directive Issuance Form

民航局 AD 編號 AD number	CAA-2020-04-002	發布日期 Date issued	2020/4/6
適用之航空產品 Applied to (models, serial numbers or part numbers, as applicable)	AS 365 N2, AS 365 N3, EC 155 B, EC 155 B1 and SA 365 N1 helicopters, all serial numbers (s/n).		
主旨摘要	Tail Rotor - Double Bearing - Inspection / Replacement		
民航局 CAA <input type="checkbox"/> 本國產品 Native products <input type="checkbox"/> 其他個案 Other	設計國民航主管機構 Original Authorities <div style="display: flex; justify-content: space-between;"> <div style="width: 48%;"> <input type="checkbox"/> FAA <input checked="" type="checkbox"/> EASA <input type="checkbox"/> Brazil <input type="checkbox"/> Transport Canada Civil Aviation <input type="checkbox"/> DGAC </div> <div style="width: 48%;"> <input type="checkbox"/> Germany LBA <input type="checkbox"/> CAA-NL <input type="checkbox"/> UK CAA <input type="checkbox"/> Japan CAB <input type="checkbox"/> CAA of Israel <input type="checkbox"/> Other _____ </div> </div>		
	設計國 AD 編號 Original AD number	2020-0078	
	1. 直接採用原 AD 之內容?(Is the original AD directly adopted?) <input checked="" type="checkbox"/> 是(Yes) <input type="checkbox"/> 否(No) _ a. 生效日期另訂為(Re-specify the effective date as) : _____ b. 執行時限另訂為(Re-specify the compliance time or period as) : _____ 2. 使用人是否需要將 AD 執行結果向民航局提出報告?(Do Users need to report the status of compliance to the CAA?) <input type="checkbox"/> 是(Yes) <input checked="" type="checkbox"/> 否(No)		
備註 Note	This AD supersedes EASA AD 2019-0267R1(CAA-2019-10-011A 修訂) dated 12 November 2019, including its Correction dated 13 November 2019.		
註： 1. AD 內容後附。 2. 航空器產品使用人得向民航局提出豁免、替代符合方法、執行時限之展延之申請。 3. 如有任何問題，請聯絡交通部民用航空局初始適航科。Tel：(02)2349-6331~3, Fax：(02)2545-8464, e-mail： adcaa@mail.caa.gov.tw Note： 1. The AD text is enclosed. 2. Exemption, an alternative method of compliance or adjustment of the compliance time may be proposed to the CAA for approval. 3. For further information, please contact Civil Aeronautics Administration on Tel：(02)2349-6331~3, Fax：(02)2545-8464, e-mail： adcaa@mail.caa.gov.tw			



Airworthiness Directive

AD No.: 2020-0078

Issued: 31 March 2020

Note: This Airworthiness Directive (AD) is issued by EASA, acting in accordance with Regulation (EU) 2018/1139 on behalf of the European Union, its Member States and of the European third countries that participate in the activities of EASA under Article 129 of that Regulation.

This AD is issued in accordance with Regulation (EU) 748/2012, Part 21.A.3B. In accordance with Regulation (EU) 1321/2014 Annex I, Part M.A.301, the continuing airworthiness of an aircraft shall be ensured by accomplishing any applicable ADs. Consequently, no person may operate an aircraft to which an AD applies, except in accordance with the requirements of that AD, unless otherwise specified by the Agency [Regulation (EU) 1321/2014 Annex I, Part M.A.303] or agreed with the Authority of the State of Registry [Regulation (EU) 2018/1139, Article 71 exemption].

Design Approval Holder's Name:

AIRBUS HELICOPTERS

Type/Model designation(s):

SA 365, AS 365 and EC 155 helicopters

Effective Date: 14 April 2020

TCDS Number(s): EASA.R.105

Foreign AD: Not applicable

Supersedure: This AD supersedes EASA AD 2019-0267R1 dated 12 November 2019, including its Correction dated 13 November 2019.

ATA 65 – Tail Rotor – Double Bearing – Inspection / Replacement

Manufacturer(s):

Airbus Helicopters (AH), formerly Eurocopter, Eurocopter France, Aerospatiale, Sud Aviation

Applicability:

AS 365 N2, AS 365 N3, EC 155 B, EC 155 B1 and SA 365 N1 helicopters, all serial numbers (s/n).

Definitions:

For the purpose of this AD, the following definitions apply:

The applicable ASB: AH AS365 Emergency Alert Service Bulletin (ASB) 01.00.71 and EC155 Emergency ASB 04A016 (single publication), as applicable.

Affected part: Double bearings, having Part Number (P/N) 704A33-651-245 or P/N 704A33-651-246, installed on a tail rotor gearbox (TGB) P/N 365A33-6005-09.

Serviceable part: Double bearings, having P/N 704A33-651-245 or P/N 704A33-651-246, which are new (never previously installed).

Groups: Group 1 helicopters are those that have an affected part installed, which has accumulated less than 500 flight hours (FH) since first installation on a helicopter.



Group 2 helicopters are those that have an affected part installed, which has accumulated 500 FH or more since first installation on a helicopter.

Close monitoring: Close monitoring procedure in accordance with Work Card 05-53-00-201 (MET) or Aircraft Maintenance Manual (AMM) Task 05-50-01-211.

Reason:

Following a regular TGB chip detector check on an AS 365 N2 helicopter, equipped with a TGB P/N 365A33-6005-09 (terminating action for EASA AD 2017-0125), a certain quantity of abrasion particles and non-critical scale particles was found, triggering the close monitoring procedure according to the applicable maintenance instructions. Since no anomaly was reported during the close monitoring regime, the helicopter returned to the normal inspection program, but after a few flight hours, during an unscheduled check, a large amount of critical scale particles was found on the chip detector's plug. The particles belonged to the double bearing (pitch control rod bearing) installed inside the TGB. The reported event showed a speed of degradation faster than expected.

This condition, if not detected and corrected, could lead to loss of yaw control of the helicopter.

To address this unsafe condition, AH issued the applicable ASB (later revised), providing inspection and replacement instructions, and EASA issued Emergency AD 2019-0267-E (also later revised) to require repetitive inspections of the TGB chip detector for particles and replacement of the double bearing.

Since EASA AD 2019-0267R1 was issued, it was determined that the interval for the chip detector inspection needs to be reduced, and AH issued the applicable ASB at Revision 2.

For the reasons described above, this AD retains the requirements of EASA AD 2019-0267R1, which is superseded, and requires the chip detector inspection for Group 1 helicopters at reduced intervals.

This AD is still considered an interim action and further AD action may follow.

Required Action(s) and Compliance Time(s):

Required as indicated, unless accomplished previously:

Inspection(s):

- (1) For Group 1 helicopters under close monitoring on 28 October 2019 [the effective date of the original issue of EASA AD 2019-0267]: Within the applicable compliance time as identified in the close monitoring and until completion of the close monitoring, assess any particles collected during the close monitoring in accordance with the instructions of paragraph 3.B.2 of the applicable ASB.
- (2) For Group 1 helicopters under close monitoring on 28 October 2019 [the effective date of the original issue of EASA AD 2019-0267]: Upon completion of the close monitoring regime, and, thereafter, within the compliance times and thereafter at intervals as specified in Table 1 of this AD, accomplish a TGB chip detector inspection in accordance with the instructions of paragraph 3.B.2 of the applicable ASB.



Table 1 – TGB Chip Detector Inspection (Monitored)

FH accumulated by double bearing (see Note 1 of this AD)	Compliance Time (whichever occurs first)	Interval (not to exceed)
Less than 250 FH	Within 20 FH after the effective date of this AD, or within 25 FH after the last inspection	20 FH
250 FH or more	Within 10 FH after the effective date of this AD, or within 25 FH after the last inspection	10 FH

Note 1: The FH are those accumulated by the double bearing on the effective date of this AD since first installation on a helicopter. For a double bearing which, after the effective date of this AD, exceeds 250 FH, the inspection interval is reduced to 10 FH.

- (3) For Group 1 helicopters not under close monitoring on 28 October 2019 [the effective date of the original issue of EASA AD 2019-0267]: Within the compliance time and, thereafter, at intervals as specified in Table 2 of this AD, accomplish a TGB chip detector inspection in accordance with the instructions of paragraph 3.B.2 of the applicable ASB.

Table 2 – TGB Chip Detector Inspection (not monitored)

FH accumulated by double bearing (see Note 1 of this AD)	Compliance Time (whichever occurs first)	Interval (not to exceed)
Less than 250 FH	Within 10 FH after the effective date of this AD, or within 25 FH after the last inspection	20 FH
250 FH or more	Within 10 FH after the effective date of this AD, or within 25 FH after the last inspection	10 FH

Assessment / Double Bearing Washing:

- (4) For Group 2 helicopters under close monitoring on 28 October 2019 [the effective date of the original issue of EASA AD 2019-0267]: Before next flight after 19 November 2019 [the effective date of EASA AD 2019-0267R1], assess any particles collected during the close monitoring in accordance with the instructions of paragraph 3.B.2 of the applicable ASB, or accomplish a double bearing washing in accordance with the instructions of paragraph 3.B.3 of the applicable ASB.
- (5) For Group 2 helicopters (all), unless already accomplished as required by paragraph (4) of this AD, within 15 FH after 28 October 2019 [the effective date of the original issue of EASA AD 2019-0267], accomplish a double bearing washing in accordance with the instructions of paragraph 3.B.3 of the applicable ASB.

Inspection(s):

- (6) For Group 2 helicopters: After accomplishment of the double bearing washing as specified in paragraph (4), or as required by paragraph (5) of this AD, as applicable, accomplish the inspections as required by paragraph (6.1) or (6.2) of this AD (see Note 2 of this AD).



- (6.1) Inspect the TGB chip detector in accordance with the instructions of paragraph 3.B.2 of the applicable ASB at intervals not exceeding 10 FH **and** accomplish a double bearing washing in accordance with the instructions of paragraph 3.B.3 of the applicable ASB at intervals not exceeding 30 FH.
- (6.2) Inspect the TGB chip detector in accordance with the instructions of paragraph 3.B.2 of the applicable ASB after every last flight of the day or at intervals not exceeding 5 FH, whichever occurs first.

Note 2: After accomplishment of a double bearing washing in accordance with the instructions of paragraph 3.B.3 of the applicable ASB, subsequent chip detector inspections can be accomplished as required by paragraph (6.1) or (6.2) of this AD.

Corrective Action(s):

- (7) If, during any action as required by paragraphs (1) to (6) of this AD, as applicable, any discrepancy, as identified in the applicable ASB, is detected, accomplish the applicable corrective action(s) in accordance with the instructions of, and within the compliance time as identified in, the applicable ASB.

Part(s) Replacement:

- (8) For all helicopters: Before an affected part exceeds 610 FH since first installation on a helicopter, or within 110 FH after 28 October 2019 [the effective date of the original issue of EASA AD 2019-0267], whichever occurs later, and, thereafter, at intervals not to exceed 500 FH, replace that affected part with a serviceable part. This can be accomplished in accordance with the instructions of the applicable AMM (see Note 3 of this AD).

Note 3: When an affected part installed on a Group 1 helicopter exceeds 500 FH since first installation on a helicopter, that helicopter effectively becomes Group 2, for which relevant requirements of this AD are then applicable.

Terminating Action(s):

- (9) None.

Parts Installation:

- (10) From 28 October 2019 [the effective date of the original issue of EASA AD 2019-0267], it is allowed to install on any helicopter a double bearing, provided it is a serviceable part, as defined in this AD, and that, following installation, it is inspected as required by this AD for Group 1 helicopters.
- (11) From 28 October 2019 [the effective date of the original issue of EASA AD 2019-0267], it is allowed to install on any helicopter a TGB P/N 365A33-6005-09, provided it is equipped with a serviceable part, as defined in this AD, and that, following installation, that serviceable part is inspected as required by this AD for Group 1 helicopters.

Ref. Publications:

AH AS365 ASB 01.00.71 original issue dated 24 October 2019, or Revision 1 dated 11 December 2019, or Revision 2 dated 25 March 2020.



AH EC155 ASB 04A016 original issue dated 24 October 2019, or Revision 1 dated 11 December 2019, or Revision 2 dated 25 March 2020.

The use of later approved revisions of the above-mentioned documents is acceptable for compliance with the requirements of this AD.

Remarks:

1. If requested and appropriately substantiated, EASA can approve Alternative Methods of Compliance for this AD.
2. Based on the required actions and the compliance time, EASA have decided to issue a Final AD with Request for Comments, postponing the public consultation process until after publication.
3. Enquiries regarding this AD should be referred to the EASA Programming and Continued Airworthiness Information Section, Certification Directorate. E-mail: ADs@easa.europa.eu.
4. Information about any failures, malfunctions, defects or other occurrences, which may be similar to the unsafe condition addressed by this AD, and which may occur, or have occurred on a product, part or appliance not affected by this AD, can be reported to the [EU aviation safety reporting system](#).
5. For any question concerning the technical content of the requirements in this AD, please contact: Airbus Helicopters (Technical Support) – Aéroport de Marseille Provence, 13725 Marignane Cedex, France, Telephone: +33 (4) 42 85 97 97, Fax: +33 (4) 42 85 99 66, Web portal: <https://keycopter.airbushelicopters.com> Technical Requests Management, or TechnicalSupport.Helicopters@airbus.com.

