



**適航指令發布單**  
**Airworthiness Directive Issuance Form**

民航局 AD 編號 AD number	CAA-2018-10-021A	發布日期 Date issued	2018/12/6												
適用之航空產品 Applied to (models, serial numbers or part numbers, as applicable)	Airbus A319-111, A319-112, A319-113, A319-114, A319-115, A319-131, A319-132 and A319-133, aeroplanes, all manufacturer serial numbers, except those on which Airbus modification (mod) 161306 has been embodied in production, and except A319 CJ aeroplanes that have not embodied Airbus mod 160001 in production, and Airbus Service Bulletin (SB) A320-57-1193 (mod 160080) in service. Airbus A320-211, A320-212, A320-214, A320-215, A320-216, A320-231, A320-232, A320-233, A321-111, A321-112, A321-131, A321-211, A321-212, A321-213, A321-231 and A321-232 aeroplanes, all manufacturer serial numbers, except those on which Airbus mod 161271 has been embodied in production.														
主旨摘要	Fuselage - Cargo Compartment Fitting Brackets, Tack and Rivet Holes - Inspection / Repair														
民航局 CAA <input type="checkbox"/> 本國產品 Native products  <input type="checkbox"/> 其他個案 Other	<div style="text-align: center;">設計國民航主管機構 Original Authorities</div> <table style="width: 100%;"><tr><td><input type="checkbox"/> FAA</td><td><input type="checkbox"/> Germany LBA</td></tr><tr><td><input checked="" type="checkbox"/> EASA</td><td><input type="checkbox"/> CAA-NL</td></tr><tr><td><input type="checkbox"/> Brazil</td><td><input type="checkbox"/> UK CAA</td></tr><tr><td><input type="checkbox"/> Transport Canada Civil Aviation</td><td><input type="checkbox"/> Japan CAB</td></tr><tr><td><input type="checkbox"/> DGAC</td><td><input type="checkbox"/> CAA of Israel</td></tr><tr><td colspan="2"><input type="checkbox"/> Other _____</td></tr></table>			<input type="checkbox"/> FAA	<input type="checkbox"/> Germany LBA	<input checked="" type="checkbox"/> EASA	<input type="checkbox"/> CAA-NL	<input type="checkbox"/> Brazil	<input type="checkbox"/> UK CAA	<input type="checkbox"/> Transport Canada Civil Aviation	<input type="checkbox"/> Japan CAB	<input type="checkbox"/> DGAC	<input type="checkbox"/> CAA of Israel	<input type="checkbox"/> Other _____	
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	設計國 AD 編號 Original AD number	2018-0233R1													
	1. 直接採用原 AD 之內容?(Is the original AD directly adopted?) <input checked="" type="checkbox"/> 是(Yes) <input type="checkbox"/> 否(No) _ a. 生效日期另訂為(Re-specify the effective date as) : _____ b. 執行時限另訂為(Re-specify the compliance time or period as) : _____														
	2. 使用人是否需要將 AD 執行結果向民航局提出報告?(Do Users need to report the status of compliance to the CAA?) <input type="checkbox"/> 是(Yes) <input checked="" type="checkbox"/> 否(No)														
備註 Note	ATA 53. This AD revises EASA AD 2018-0233(CAA-2018-10-021) dated 26 October 2018, which superseded EASA AD 2013-0310(CAA-2013-12-015) dated 20 December 2013. Ref. Publications: Airbus SB A320-53-1257 original issue dated 21 December 2012, or Rev. 01 dated 28 April 2016, or Rev. 02 dated 29 November 2016. and Airbus SB A320-53-1261 original issue dated 21 December 2012, or Rev. 01 dated 30 June 2015, or Rev. 02 dated 01 March 2016, or Rev. 03 dated 08 August 2016, or Rev. 04 dated 03 May 2017. and Airbus SB A320-53-1360 original issue dated 01 December 2016, or Rev. 01 dated 01 March 2018. and Airbus SB A320-53-1364 original issue dated 04 May 2018. Airbus SB A320-53-1365 original issue dated 17 April 2018.														
註： 1. AD 內容後附。 2. 航空器產品使用人得向民航局提出豁免、替代符合方法、執行時限之展延之申請。 3. 如有任何問題，請聯絡交通部民用航空局初始適航科。Tel：(02)2349-6331~3, Fax：(02)2545-8464, e-mail： <a href="mailto:adcaa@mail.caa.gov.tw">adcaa@mail.caa.gov.tw</a> Note： 1. The AD text is enclosed. 2. Exemption, an alternative method of compliance or adjustment of the compliance time may be proposed to the CAA for approval. 3. For further information, please contact Civil Aeronautics Administration on Tel：(02)2349-6331~3, Fax：(02)2545-8464, e-mail： <a href="mailto:adcaa@mail.caa.gov.tw">adcaa@mail.caa.gov.tw</a>															





## Airworthiness Directive

**AD No.:** 2018-0233R1

**Issued:** 28 November 2018

Note: This Airworthiness Directive (AD) is issued by EASA, acting in accordance with Regulation (EU) 2018/1139 on behalf of the European Union, its Member States and of the European third countries that participate in the activities of EASA under Article 129 of that Regulation.

This AD is issued in accordance with Regulation (EU) 748/2012, Part 21.A.3B. In accordance with Regulation (EU) 1321/2014 Annex I, Part M.A.301, the continuing airworthiness of an aircraft shall be ensured by accomplishing any applicable ADs. Consequently, no person may operate an aircraft to which an AD applies, except in accordance with the requirements of that AD, unless otherwise specified by the Agency [Regulation (EU) 1321/2014 Annex I, Part M.A.303] or agreed with the Authority of the State of Registry [Regulation (EU) 2018/1139, Article 71 exemption].

### Design Approval Holder's Name:

AIRBUS

### Type/Model designation(s):

A319, A320 and A321 aeroplanes

**Effective Date:** Revision 01: 05 December 2018  
Original issue: 09 November 2018

**TCDS Number(s):** EASA.A.064

**Foreign AD:** Not applicable

**Supersedure:** This AD revises EASA AD 2018-0233 dated 26 October 2018, which superseded EASA AD 2013-0310 dated 20 December 2013.

## ATA 53 – Fuselage – Cargo Compartment Fitting Brackets, Tack and Rivet Holes – Inspection / Repair

### Manufacturer(s):

Airbus, formerly Airbus Industrie

### Applicability:

Airbus A319-111, A319-112, A319-113, A319-114, A319-115, A319-131, A319-132 and A319-133, aeroplanes, all manufacturer serial numbers, except those on which Airbus modification (mod) 161306 has been embodied in production, and except A319 CJ aeroplanes that have not embodied Airbus mod 160001 in production, and Airbus Service Bulletin (SB) A320-57-1193 (mod 160080) in service.

Airbus A320-211, A320-212, A320-214, A320-215, A320-216, A320-231, A320-232, A320-233, A321-111, A321-112, A321-131, A321-211, A321-212, A321-213, A321-231 and A321-232 aeroplanes, all manufacturer serial numbers, except those on which Airbus mod 161271 has been embodied in production.

### Definitions:

For the purpose of this AD, the following definitions apply:



**The inspection SB:** Airbus SB A320-53-1257 Revision (Rev.) 02.

**The applicable modification SB:** Airbus SB A320-53-1261 Rev. 04, SB A320-53-1360 (any revision), SB A320-53-1364 and SB A320-53-1365, as applicable.

**A319 CJ:** A319 (Corporate Jet) aeroplanes on which Airbus mod 28238, mod 28162 and mod 28342 have been embodied in production.

**A319 PAX:** A319 aeroplanes, which are not A319 CJ.

#### Reason:

During a full scale fatigue test, several broken frames in the cargo compartment area between Frame (FR) 50 and FR63 have been found, especially on the cargo floor support fittings and open tack holes on left hand (LH) side.

This condition, if not detected and corrected, could affect the structural integrity of the aeroplane.

To address this unsafe condition, Airbus issued SB A320-53-1257, providing inspection instructions, and SB A320-53-1261, providing modification instructions.

Consequently, EASA published AD 2013-0310, requiring repetitive inspections of the frames in the cargo compartment area and of the cargo floor support fittings and open tack holes on the LH side and, depending on findings, accomplishment of corrective action(s). That AD also required a modification, which constituted terminating action for the required repetitive inspections.

After that AD was issued, further analyses and widespread fatigue damage evaluations identified the need to reduce the threshold and intervals for the repetitive inspections for certain configurations, and Airbus issued the inspection SB accordingly. Airbus issued SB A320-53-1360, SB A320-53-1364 and SB A320-53-1365 to supplement SB A320-53-1261, and SB Information Transmission (SBIT) 16-0070 providing additional information. Consequently, EASA issued AD 2018-0233, retaining the requirements of EASA AD 2013-0310, which was superseded, but requiring accomplishment of the repetitive inspections within reduced compliance times for certain configurations. That AD also required additional work for aeroplanes that had already been modified in accordance with the instructions of Airbus SB A320-53-1261 Rev. 02.

Since that AD was issued, it has been determined that certain A319 aeroplanes may be excluded from the Applicability of the AD, since the calculated compliance time for the initial inspection is beyond the applicable limit of validity.

For the reason described above, this AD is revised to reduce the Applicability.

#### Required Action(s) and Compliance Time(s):

Required as indicated, unless accomplished previously:

#### Repetitive Inspection(s):

- (1) Within the compliance time(s) as defined in Table 1 or Table 2 of Appendix 1 of this AD, as applicable, and, thereafter, at intervals not to exceed the values as defined in Table 3 of



Appendix 1 of this AD, as applicable, accomplish a rototest inspection of open tack holes and rivet holes at the cargo floor support fittings between FR50 and FR63, between Stringer (STG) 33 and STG39 (LH side only), for A320 and A321 aeroplanes, and between FR53 and FR63, between STG33 and STG39 (LH side only), for A319 aeroplanes, in accordance with the instructions of the inspection SB.

#### **Corrective Action(s):**

- (2) If, during any inspection as required by paragraph (1) of this AD, a crack is detected, before next flight, contact Airbus for approved repair instructions and, within the compliance time identified in those instructions, accomplish those instructions accordingly.

#### **Modification:**

- (3) Except as required by paragraph (4) of this AD, before exceeding 48 000 flight cycles (FC) or 96 000 flight hours (FH), whichever occurs first since aeroplane first flight, modify the aeroplane in accordance with the instructions of the applicable modification SB.

#### **Additional Work:**

- (4) For an aeroplane that was modified, before the effective date of this AD, in accordance with the instructions of Airbus SB A320-53-1261 Rev. 02, within 30 days after 09 November 2018 [the effective date of the original issue of this AD], contact Airbus for additional work instructions and, within the compliance time specified in those instructions, accomplish the additional work accordingly.

#### **Credit:**

- (5) Modification of an aeroplane, before 09 November 2018 [the effective date of the original issue of this AD], in accordance with the instructions of Airbus SB A320-53-1261 original issue, or Rev. 01, or Rev. 03, is acceptable to comply with the requirements of paragraph (3) of this AD for that aeroplane.
- (6) Inspections on an aeroplane, accomplished before 09 November 2018 [the effective date of the original issue of this AD], in accordance with the instructions of Airbus SB A320-53-1257 at original issue or Rev. 01, are acceptable to comply with the initial requirements of paragraph (1) of this AD for that aeroplane.

#### **Terminating Action:**

- (7) Repair of an aeroplane as required by paragraph (2) of this AD does not constitute terminating action for the repetitive inspections as required by paragraph (1) of this AD for that aeroplane, unless otherwise stated in the repair instructions.
- (8) Modification of an aeroplane as required by paragraph (3) or (4) of this AD, as applicable, constitutes terminating action for the repetitive inspections as required by paragraph (1) of this AD for that aeroplane.

#### **Ref. Publications:**

Airbus SB A320-53-1257 original issue dated 21 December 2012, or Rev. 01 dated 28 April 2016, or Rev. 02 dated 29 November 2016.



Airbus SB A320-53-1261 original issue dated 21 December 2012, or Rev. 01 dated 30 June 2015, or Rev. 02 dated 01 March 2016, or Rev. 03 dated 08 August 2016, or Rev. 04 dated 03 May 2017.

Airbus SB A320-53-1360 original issue dated 01 December 2016, or Rev. 01 dated 01 March 2018.

Airbus SB A320-53-1364 original issue dated 04 May 2018.

Airbus SB A320-53-1365 original issue dated 17 April 2018.

The use of later approved revisions of the above-mentioned documents is acceptable for compliance with the requirements of this AD.

#### Remarks:

1. If requested and appropriately substantiated, EASA can approve Alternative Methods of Compliance for this AD.
2. The original issue of this AD was posted on 11 July 2018 as PAD 18-093 for consultation until 08 August 2018. The Comment Response Document can be found in the [EASA Safety Publications Tool](#), in the compressed (zipped) file attached to the record for this AD.
3. Enquiries regarding this AD should be referred to the EASA Safety Information Section, Certification Directorate. E-mail: [ADs@easa.europa.eu](mailto:ADs@easa.europa.eu).
4. Information about any failures, malfunctions, defects or other occurrences, which may be similar to the unsafe condition addressed by this AD, and which may occur, or have occurred on a product, part or appliance not affected by this AD, can be reported to the [EU aviation safety reporting system](#).
5. For any question concerning the technical content of the requirements in this AD, please contact: AIRBUS – Airworthiness Office – EIAS; Fax +33 5 61 93 44 51;  
E-mail: [account.airworth-eas@airbus.com](mailto:account.airworth-eas@airbus.com).



## Appendix 1

Table 1 - Initial Inspection

<b>Model and Configuration</b>	<b>FH and FC Accumulated</b> on 03 January 2014 [the effective date of EASA AD 2013-0310] since first flight of the aeroplane	<b>Compliance Time</b>
A320 and A319 PAX, pre-mod 160001 and pre-SB A320-57-1193 (mod 160080)	36 200 FC or 72 400 FH, or less	Before exceeding 38 200 FC or 76 400 FH, whichever occurs first since first flight of the aeroplane
	More than 36 200 FC or 72 400 FH, but not more than 45 000 FC	Within 2 000 FC or 4 000 FH whichever occurs first after 03 January 2014 [the effective date of AD 2013-0310]
A321 pre-mod 160021	More than 45 000 FC	Within 1 000 FC or 2 000 FH, whichever occurs first after 03 January 2014 [the effective date of AD 2013-0310]

Table 2 - Initial Inspection

<b>Model and Configuration</b>	<b>FH and FC Accumulated</b> on 09 November 2018 [the effective date of the original issue of this AD] since first flight of the aeroplane	<b>Compliance Time</b> (see Note 1 of this AD)
A320 and A319 PAX, post-mod 160001 or post-SB A320-57-1193 (mod 160080)  A321 post-mod 160021	33 400 FC or 66 900 FH, or less	Before exceeding 35 400 FC or 70 900 FH, whichever occurs first since first flight of the aeroplane (see Note 2 of this AD)
	More than 33 400 FC or 66 900 FH, but no more than 40 000 FC	Within 2 000 FC or 4 000 FH, whichever occurs first since 09 November 2018 [the effective date of the original issue of this AD] (see Note 2 of this AD)
	More than 40 000 FC but no more than 43 000 FC	Within 1 000 FC or 2 000 FH, whichever occurs first since 09 November 2018 [the effective date of the original issue of this AD] (see Note 2 of this AD)
	More than 43 000 FC	Within 30 days after 09 November 2018 [the effective date of the original issue of this AD] (see Note 2 of this AD)
A319 CJ, post-mod 160001 or post-SB A320-57-1193 (mod 160080)	Not applicable	Before exceeding 19 800 FC or 85 300 FH, whichever occurs first since first flight of the aeroplane



Note 1: For A320 and A319 in post-SB A320-57-1193 configuration, refer to Airworthiness Limitations Section (ALS) Part 2 variation 3.6 or ALS Part 2 revision 4, or later further ALS Part 2 revision, for determination of the threshold when sharklet is installed.

Note 2: For A320 and A319 PAX, post-SB A320-57-1193 (mod 160080): Without exceeding the time at which inspection is required through the threshold or compliance time for A320 and A319 PAX aeroplanes in pre-SB A320-57-1193 (pre mod 160080) configuration.

Table 3 – Repetitive Inspections

<b>Model and Configuration</b>	<b>Interval (FC or FH, whichever occurs first)</b>
A320, A319 PAX and A321	5 000 FC or 10 000 FH
A319 CJ, post mod 160001 or post-SB A320-57-1193 (mod 160080)	2 900 FC or 12 800 FH

