

 適航指令發布單 Airworthiness Directive Issuance Form			
民航局 AD 編號 AD number	CAA-2017-12-005	發布日期 Date issued	2017/12/22
適用之航空產品 Applied to (models, serial numbers or part numbers, as applicable)	Airbus A318-111, A318-112, A319-111, A319-112, A319-113, A319-114, A319-115, A320-211, A320-212, A320-214, A320-215, A320-216, A321-111, A321-112, A321-211, A321-212, and A321-213 aeroplanes, all manufacturer serial numbers.		
主旨摘要	Powerplant - Aft Engine Mount Retainers - Inspection / Replacement		
民航局 CAA <input type="checkbox"/> 本國產品 Native products <input type="checkbox"/> 其他個案 Other	設計國民航主管機構 Original Authorities <input type="checkbox"/> FAA <input type="checkbox"/> Germany LBA <input checked="" type="checkbox"/> EASA <input type="checkbox"/> CAA-NL <input type="checkbox"/> Brazil <input type="checkbox"/> UK CAA <input type="checkbox"/> Transport Canada Civil Aviation <input type="checkbox"/> Japan CAB <input type="checkbox"/> DGAC <input type="checkbox"/> CAA of Israel <input type="checkbox"/> Other _____		
	設計國 AD 編號 Original AD number	2017-0251	
	1. 直接採用原 AD 之內容?(Is the original AD directly adopted?) <input checked="" type="checkbox"/> 是(Yes) <input type="checkbox"/> 否(No) _ a. 生效日期另訂為(Re-specify the effective date as) : _____ b. 執行時限另訂為(Re-specify the compliance time or period as) : _____ 2. 使用人是否需要將 AD 執行結果向民航局提出報告?(Do Users need to report the status of compliance to the CAA?) <input type="checkbox"/> 是(Yes) <input checked="" type="checkbox"/> 否(No)		
備註 Note	ATA 71. This AD supersedes EASA AD 2017-0138(CAA-2017-08-001), dated 02 August 2017. Ref. Publications: Airbus AOT A71N001-12 Revision 01 dated 09 August 2012, or Revision 02 dated 27 February 2013. and Airbus AOT A71N011-15 original issue dated 16 September 2015, or Revision 01 dated 01 February 2016. and Airbus SB A320-71-1060 original issue dated 09 October 2014, or Revision 01 dated 07 April 2015, or Revision 02 dated 18 December 2015. and Airbus SB A320-71-1070 original issue dated 23 November 2015. and Airbus SB A320-71-1071 original issue dated 08 November 2016 or Revision 01 dated 17 October 2017.		
註：	1. AD 內容後附。 2. 航空器產品使用人得向民航局提出豁免、替代符合方法、執行時限之展延之申請。 3. 如有任何問題，請聯絡交通部民用航空局初始適航科。Tel：(02)2349-6331~3, Fax：(02)2545-8464, e-mail： adcaa@mail.caa.gov.tw		
Note：	1. The AD text is enclosed. 2. Exemption, an alternative method of compliance or adjustment of the compliance time may be proposed to the CAA for approval. 3. For further information, please contact Civil Aeronautics Administration on Tel:(02)2349-6331~3, Fax:(02)2545-8464, e-mail： adcaa@mail.caa.gov.tw		



Airworthiness Directive

AD No.: 2017-0251

Issued: 15 December 2017

Note: This Airworthiness Directive (AD) is issued by EASA, acting in accordance with Regulation (EC) 216/2008 on behalf of the European Union, its Member States and of the European third countries that participate in the activities of EASA under Article 66 of that Regulation.

This AD is issued in accordance with Regulation (EU) 748/2012, Part 21.A.3B. In accordance with Regulation (EU) 1321/2014 Annex I, Part M.A.301, the continuing airworthiness of an aircraft shall be ensured by accomplishing any applicable ADs. Consequently, no person may operate an aircraft to which an AD applies, except in accordance with the requirements of that AD, unless otherwise specified by the Agency [Regulation (EU) 1321/2014 Annex I, Part M.A.303] or agreed with the Authority of the State of Registry [Regulation (EC) 216/2008, Article 14(4) exemption].

Design Approval Holder's Name:

AIRBUS

Type/Model designation(s):

A318, A319, A320, and A321 aeroplanes

Effective Date: 29 December 2017

TCDS Number(s): EASA.A.064

Foreign AD: Not applicable

Revision: This AD supersedes EASA AD 2017-0138, dated 02 August 2017.

ATA 71 – Powerplant – Aft Engine Mount Retainers – Inspection / Replacement

Manufacturer(s):

Airbus (formerly Airbus Industrie)

Applicability:

Airbus A318-111, A318-112, A319-111, A319-112, A319-113, A319-114, A319-115, A320-211, A320-212, A320-214, A320-215, A320-216, A321-111, A321-112, A321-211, A321-212, and A321-213 aeroplanes, all manufacturer serial numbers.

Reason:

During in-service inspections, several aft engine mount inner retainers, fitted on aeroplanes equipped with CFM56-5A/5B engines, were found broken. Investigation identified that the main cause of crack initiation was the vibration dynamic effect that affects the retainers, and that the “dull” surface finish pitting is an aggravating factor when compared with the “bright” surface finishing.

This condition, if not detected and corrected, could lead to in-flight loss of an aft engine mount link, possibly resulting in damage to the aeroplane and/or injury to persons on the ground.

To address this potential unsafe condition, Airbus issued Alert Operators Transmission (AOT) A71N001-12 (later revised) and EASA issued AD 2013-0050, later superseded by EASA



AD 2015-0021, requiring repetitive detailed inspections (DET) of all aft engine mount inner retainers and, depending on findings, their replacement.

After EASA AD 2015-0021 was issued, a production quality deficiency was identified by Airbus and Goodrich Aerostructures, the engine mount retainer manufacturer, on the inner retainer, Part Number (P/N) 238-0252-505, installed in the three link assemblies of the engine mount fitted on CFM56-5A/5B engines. Airbus issued AOT A71N011-15 and Service Bulletin (SB) A320-71-1070, providing a list of affected parts and applicable corrective actions.

Consequently, EASA issued AD 2016-0010 (later revised), retaining the requirements of EASA AD 2015-0021, which was superseded, and in addition requiring the identification and replacement of all non-conforming aft engine mount inner retainers.

After that AD was issued, a new engine mount retainer was developed by Goodrich Aerostructures to improve the retainer efficiency. For retrofit purposes, Goodrich Aerostructures issued SB RA32071-164, and Airbus issued SB A320-71-1071, providing instructions to modify and re-identify the engine mount assemblies as instructed in the Goodrich Aerostructures SB. Subsequently, it was observed that, on aeroplanes equipped with certain engines fitted with a Turbine Rear Frame (TRF) with 4 lugs configuration, the installation of the new engine mount retainers can lead to interference, and Goodrich Aerostructures revised SB RA32071-164, providing instructions not to install the new engine retainers on affected engines. For engines fitted with a TRF with 4 lugs, a new installation (potentially requiring different engine mount retainers) is being developed by Goodrich Aerospace and Airbus.

Consequently, EASA issued AD 2017-0138, retaining the requirements of EASA AD 2016-0010R1, which was superseded, and, except for aeroplanes equipped with engines fitted with a TRF with 4 lugs configuration, requiring modification and identification of aft engine mount assemblies as terminating action for the repetitive inspections of the retainers. That AD also included additional instructions applicable to installation of engines fitted with a TRF with 4 lugs configuration.

Since EASA AD 2017-0138 was issued, it was determined that installation of new engine mount assemblies must not be allowed for some specific engine configurations, and that installation of Goodrich Aerostructures SB RA32071-164 alone can be referred to, in order to accomplish the terminating action as required by that AD.

For the reason described above, this AD retains the requirement of EASA AD 2017-0138, which is superseded, adds reference to Goodrich Aerostructures SB RA32071-164 in paragraph (9), and introduces new requirement for aeroplanes equipped with engines fitted with a TRF with 4 lugs configuration.

Required Action(s) and Compliance Time(s):

Required as indicated, unless accomplished previously:

Restatement of the requirements of EASA AD 2017-0138:

Note 1: Group 1 aeroplanes have an aft engine mount assembly installed, having a P/N identified as "Old" in Table 1 of this AD. Group 2 aeroplanes do not have an aft engine mount assembly installed having a P/N identified as "Old" in Table 1 of this AD.



Note 2: For the purpose of this AD, a “4-lugs engine” is a CFM56-5A1, CFM56-5A3, CFM56-5A4, CFM56-5A4/F, CFM56-5A5 or CFM56-5A5/F engine, fitted with a TRF having a P/N as identified in Appendix 1 of this AD.

Table 1: Aft Engine Mount Re-identification P/N

Old P/N	New P/N
238-0230-11	238M0230-11
238-0230-15	238M0230-15
238-0230-5	238M0230-5
642-2300-3	642-2300-11

One-time Inspection:

- (1) For Group 1 aeroplanes (see Note 1 of this AD): Within 3 months after 19 March 2013 [the effective date of EASA AD 2013-0050], accomplish a DET of the aft engine mount inner retainers in accordance with the instructions of Airbus AOT A71N001-12 Revision (Rev.) 01.

Aeroplanes from MSN 4942 and higher have been delivered by Airbus with “bright” finish aft engine mount retainers. These aeroplanes are not affected by the requirements of paragraph (1) of this AD, provided it is determined that no engine and/or no aft engine mount inner retainer has been removed from the aeroplane since Airbus date of manufacture.

Repetitive Inspections:

- (2) For Group 1 aeroplanes (see Note 1 of this AD): Within the compliance time as specified in Table 2 of this AD and, thereafter, at intervals not to exceed 12 months, accomplish a DET of the aft engine mount inner retainers in accordance with the instructions of Airbus SB A320-71-1060, or Goodrich Aerostructures SB RA32071-160.

Table 2 – Inspection Threshold

Compliance Time (A, B or C, whichever occurs later)	
A	Within 12 months since Airbus date of manufacture of the aeroplane
B	Within 12 months after installation of new inner retainers
C	Within 9 months after 27 February 2015 [the effective date of EASA AD 2015-0021]

Corrective Action(s):

- (3) If, during the DET as required by paragraph (1) of this AD, any installed “dull” finish aft engine mount inner retainer is found without cracks and not failed, within 25 flight cycles (FC), repeat the DET as required by paragraph (1) of this AD and, within 50 FC after the first DET as required by paragraph (1) of this AD, replace all “dull” finish inner retainers in accordance with the instructions of Airbus AOT A71N001-12 Rev. 01.



- (4) If, during the DET as required by paragraph (1) of this AD, any installed aft engine mount inner retainer is found cracked or failed, before next flight, replace all affected aft engine mount inner retainers in accordance with the instructions of Airbus AOT A71N001-12 Rev. 01.
- (5) If, during any DET as required by paragraph (2) of this AD, any aft engine mount inner retainer is found damaged, cracked or broken, or detected as missing, before next flight, replace the affected aft engine mount inner retainers of the affected engine installation in accordance with the instructions of Airbus SB A320-71-1060.

Part Identification / Replacement:

- (6) For Group 1 aeroplanes (see Note 1 of this AD): Within 2 months after 27 January 2016 [the effective date of EASA AD 2016-0010], identify each engine mount inner retainer in accordance with the instructions of Airbus SB A320-71-1070, and replace each part that meets any of the criteria as specified in paragraph (6.1), (6.2) or (6.3) of this AD, as applicable.

(6.1) Part listed in Table 1 of Airbus AOT A71N011-15 Rev. 01.

(6.2) Part installed since the aeroplane date of manufacture, or since 01 March 2015 (whichever occurred later) and before 27 January 2016 [the effective date of the original issue of EASA AD 2016-0010], which can be identified by a Purchase Order (PO) as listed in Table 2 of Airbus AOT A71N011-15 Rev. 01.

(6.3) Part installed since the aeroplane date of manufacture, or since 01 March 2015 (whichever occurred later) and before 27 January 2016 [the effective date of the original issue of EASA AD 2016-0010], which cannot be identified by a PO.

The use of Airbus AOT A71N011-15 Rev. 01 or Goodrich Aerostructures SB RA32071-165 is an acceptable method in lieu of Airbus SB A320-71-1070 to comply with paragraph (6) of this AD.

A review of aeroplane maintenance records is acceptable to make this identification, provided those records can be relied upon for the purpose of this requirement.

Parts Installation:

- (7) From 19 March 2013 [the effective date of EASA AD 2013-0050], do not install on any aeroplane a “dull” finish aft engine mount inner retainer. The instructions of Airbus AOT A71N001-12, or Goodrich SB RA32071-146, can be used to verify the correct finish of the part.
- (8) From 27 January 2016 [the effective date of the original issue of EASA AD 2016-0010], do not install on any aeroplane an engine mount inner retainer that meets any of the criteria as specified in paragraph (8.1), (8.2) or (8.3) of this AD, as applicable.

(8.1) Part delivered through a PO as listed in Table 2 of AOT A71N011-15 Rev. 01.

(8.2) Part delivered through an unidentified PO.

(8.3) Part listed in Table 1 of AOT A71N011-15 Rev. 01.



Modification:

- (9) For Group 1 aeroplanes (see Note 1 of this AD): Within 48 months after 16 August 2017 [the effective date of EASA AD 2017-0138], except for “4-lugs engine” (see Note 2 of this AD), modify the aft engine mount assembly, having a P/N identified as “Old” in Table 1 of this AD, and re-identify it with the corresponding P/N identified as “New” in Table 1 of this AD, in accordance with the instructions of Airbus SB A320-71-1071, or Goodrich Aerostructures SB RA32071-164.

Alternative Method:

- (10) Replacement on an aeroplane of each aft engine mount assembly, identified as “Old P/N” in Table 1 of this AD, with a corresponding aft engine mount assembly, identified as “New P/N” in Table 1 of this AD, is an acceptable method to comply with the requirements of paragraph (9) of this AD for that aeroplane.

Credit:

- (11) An aeroplane on which Airbus modification 158435 has been embodied in production is a Group 2 aeroplane (see Note 1 of this AD), provided that it is determined that no aft engine mount assembly, identified as “Old P/N” in Table 1 of this AD, has been installed on that aeroplane after the aeroplane date of manufacture.

A review of aeroplane maintenance records is acceptable to make this determination, provided those records can be relied upon for that purpose.

Parts Installation:

- (12) Do not install an aft engine mount assembly identified as “Old P/N” in Table 1 of this AD on any aeroplane, as required by paragraph (12.1) or (12.2) of this AD, as applicable (see Note 1 of this AD).
- (12.1) For Group 1 aeroplanes: After modification of the aeroplane as required by paragraph (9) of this AD, or as specified in paragraph (10) of this AD, as applicable.
- (12.2) For Group 2 aeroplanes: From 16 August 2017 [the effective date of EASA AD 2017-0138].
- (13) From 16 August 2017 [the effective date of EASA AD 2017-0138], it is allowed to (re)install a “4 lugs engine” on an aeroplane (left-hand (LH) and/or right-hand (RH) side) provided that the aeroplane is equipped with an aft engine mount assembly identified as “Old P/N” in Table 1 of this AD on the affected engine pylon (LH and/or RH).

Terminating Action:

- (14) Replacement of inner retainers on an aeroplane as required by paragraph (5) or (6) of this AD, as applicable, does not constitute terminating action for the repetitive DET as required by paragraph (2) of this AD for that aeroplane.
- (15) Modification of an aeroplane as required by paragraph (9) of this AD, or as specified in paragraph (10) of this AD, as applicable, constitutes terminating action for the repetitive DET



as required by paragraph (2) for that aeroplane, and compliance with the requirements of paragraph (6) of this AD for that aeroplane.

New Requirements of this AD:

Parts Installation:

- (16) For aeroplanes equipped with a “4 lugs engine” (LH and/or RH side): From the effective date of this AD, do not modify, as required by paragraph (9) of this AD, any aft engine mount assembly (LH and/or RH), having a P/N identified as “Old” in Table 1 of this AD, and do not install on the affected engine pylon (LH and/or RH) any aft engine mount assembly, having a P/N identified as “New” in Table 1 of this AD.
- (17) For an aeroplane equipped with a “4 lugs engine” (LH and/or RH side), and on which, before the effective date of this AD, an aft engine mount assembly, having a P/N identified as “New” in Table 1 of this AD has been installed on the affected engine pylon (LH and/or RH), or on which the aft engine mount assembly (LH and/or RH), having a P/N identified as “Old” in Table 1 of this AD, has been modified as required by paragraph (9) of this AD, before next flight after the effective date of this AD, contact Airbus for approved instructions and accomplish those instructions accordingly.

Ref. Publications:

Airbus AOT A71N001-12 Revision 01 dated 09 August 2012, or Revision 02 dated 27 February 2013.

Airbus AOT A71N011-15 original issue dated 16 September 2015, or Revision 01 dated 01 February 2016.

Airbus SB A320-71-1060 original issue dated 09 October 2014, or Revision 01 dated 07 April 2015, or Revision 02 dated 18 December 2015.

Airbus SB A320-71-1070 original issue dated 23 November 2015.

Airbus SB A320-71-1071 original issue dated 08 November 2016 or Revision 01 dated 17 October 2017.

Goodrich Aerostructures SB RA32071-146 Revision 02 dated 26 July 2012.

Goodrich Aerostructures SB RA32071-160 original issue dated 18 September 2014.

Goodrich Aerostructures SB RA32071-165 original issue dated 09 October 2015.

Goodrich Aerostructures SB RA32071-164 original issue dated 06 October 2016, or Revision 01 dated 19 July 2017.

The use of later approved revisions of these documents is acceptable for compliance with the requirements of this AD.



Remarks:

1. If requested and appropriately substantiated, EASA can approve Alternative Methods of Compliance for this AD.
2. Based on the required actions and the compliance time, EASA have decided to issue a Final AD with Request for Comments, postponing the public consultation process until after publication.
3. Enquiries regarding this AD should be referred to the EASA Safety Information Section, Certification Directorate. E-mail: ADs@easa.europa.eu.
4. For any question concerning the technical content of the requirements in this AD, please contact: AIRBUS – Airworthiness Office – EIAS; Fax +33 5 61 93 44 51;
E-mail: account.airworth-eas@airbus.com.



Appendix 1 – TRF with 4 lugs configuration

P/N
336-031-615-0
336-031-617-0
336-031-618-0
336-031-621-0
336-031-650-0
336-031-651-0
336-031-652-0
336-031-653-0
336-031-660-0
336-031-661-0
336-031-662-0
336-031-663-0
336-031-670-0
336-031-671-0
336-031-672-0
336-031-673-0
336-031-640-0
336-031-642-0

