

 <b>適航指令發布單</b> <b>Airworthiness Directive Issuance Form</b>			
民航局AD編號 AD Number	CAA-2016-06-005	發布日期 Date issued	2016/6/13
適用之航空產品 Applied to (models, serial numbers or part numbers, as applicable)	Airbus A321-111, A321-112, A321-131, A321-211, A321-212, A321-213, A321-231 and A321-232 aeroplanes, all manufacturer serial numbers.		
主旨摘要	Fuselage – Frame 35.2A - Inspection		
民航局 CAA  <input type="checkbox"/> 本國產品 Native products  <input type="checkbox"/> 其他個案 Other	設計國民航主關機構 Original Authorities <div style="display: flex; justify-content: space-between;"> <div> <input type="checkbox"/> FAA  <input checked="" type="checkbox"/> EASA  <input type="checkbox"/> Brazil  <input type="checkbox"/> Transport Canada Civil Aviation  <input type="checkbox"/> DGAC               </div> <div> <input type="checkbox"/> Germany LBA  <input type="checkbox"/> CAA-NL  <input type="checkbox"/> UK CAA  <input type="checkbox"/> Japan CAB  <input type="checkbox"/> CAA of Israel  <input type="checkbox"/> Other _____               </div> </div>		
	設計國AD編號 Original AD number	2016-0106	
	1. 直接採用原AD之內容?(Is the original AD directly adopted?) <input checked="" type="checkbox"/> 是(Yes) <input type="checkbox"/> 否(No) ____ a. 生效日期另訂為(Re-specify the effective date as) : _____ b. 執行時限另訂為(Re-specify the compliance time or period as) : _____ 2. 使用人是否需要將AD執行結果向民航局提出報告?(Do Users need to report the status of compliance to the CAA?) <input type="checkbox"/> 是(Yes) <input checked="" type="checkbox"/> 否(No)		
備註 Note	Airbus SB A320-53-1315 original issue dated 13 January 2016. Airbus SB A320-53-1316 original issue dated 13 January 2016. The use of later approved revisions of these documents is acceptable for compliance with the requirements of this AD.		
註： 1. AD內容後附。 2. 航空器產品使用人得向民航局提出豁免、替代符合方法、執行時限之展延之申請。 3. 如有任何問題，請聯絡交通部民用航空局初始適航科。Tel：(02)2349-6331~3, Fax：(02)2545-8464, e-mail：adcaa@mail.caa.gov.tw Note： 1. The AD text is enclosed. 2. Exemption, an alternative method of compliance or adjustment of the compliance time may be proposed to the CAA for approval. 3. For further information, please contact Civil Aeronautics Administration on Tel：(02)2349-6331~3, Fax：(02)2545-8464, e-mail：adcaa@mail.caa.gov.tw			



## Airworthiness Directive

**AD No.:** 2016-0106

**Issued:** 06 June 2016

Note: This Airworthiness Directive (AD) is issued by EASA, acting in accordance with Regulation (EC) 216/2008 on behalf of the European Union, its Member States and of the European third countries that participate in the activities of EASA under Article 66 of that Regulation.

This AD is issued in accordance with Regulation (EU) 748/2012, Part 21.A.3B. In accordance with Regulation (EU) 1321/2014 Annex I, Part M.A.301, the continuing airworthiness of an aircraft shall be ensured by accomplishing any applicable ADs. Consequently, no person may operate an aircraft to which an AD applies, except in accordance with the requirements of that AD, unless otherwise specified by the Agency [Regulation (EU) 1321/2014 Annex I, Part M.A.303] or agreed with the Authority of the State of Registry [Regulation (EC) 216/2008, Article 14(4) exemption].

**Design Approval Holder's Name:**

AIRBUS

**Type/Model designation(s):**

A321 aeroplanes

**Effective Date:** 13 June 2016

**TCDS Number(s):** EASA.A.064

**Foreign AD:** Not applicable

**Supersedure:** None

### ATA 53 – Fuselage – Frame 35.2A - Inspection

#### Manufacturer(s):

Airbus (formerly Airbus Industrie)

#### Applicability:

Airbus A321-111, A321-112, A321-131, A321-211, A321-212, A321-213, A321-231 and A321-232 aeroplanes, all manufacturer serial numbers.

#### Reason:

Following a new full scale fatigue test campaign on the A321 airframe, in the context of the A321 extended service goal, it was identified that cracks could develop on holes at frame (FR) 35.2A between stringers (STR) 22 and STR 23 on right hand (RH) and left hand (LH) sides, also on aeroplanes operated in the context of design service goal.

This condition, if not detected and corrected, could reduce the structural integrity of the fuselage.

Prompted by these findings, Airbus developed an inspection programme, published in Service Bulletin (SB) A320-53-1315 and SB A320-53-1316, each containing instructions for a different location.

For the reasons described above, this AD requires repetitive special detailed (rototest) inspections (SDI) of the affected holes and, depending on findings, accomplishment of a repair.



This AD is considered an interim action, pending development of a permanent solution.

**Required Action(s) and Compliance Time(s):**

Required as indicated, unless accomplished previously:

- (1) Within the compliance time specified in Table 1 of this AD, as applicable, and, thereafter, at intervals not to exceed 21 500 flight cycles (FC) or 43 100 flight hours (FH), whichever occurs first, accomplish a SDI of the affected holes at FR35.2A between STR22 and STR23 at the locations as specified in Table 2 of this AD and in accordance with the instructions of the applicable Airbus SB as defined in Table 2 of this AD.

Table 1 – Inspection Threshold

<b>Compliance Time (whichever occurs later, A or B)</b>	
<b>A</b>	Before exceeding 25 400 FC or 50 900 FH, whichever occurs first since aeroplane first flight
<b>B</b>	Within 3 300 FC after the effective date of this AD

Table 2 – Inspection Locations and Applicable SB

<b>Location</b>	<b>SB</b>
FR 35.2A RH side	A320-53-1315
FR 35.2A LH side	A320-53-1316

- (2) If, during any SDI as required by paragraph (1) of this AD, any crack is found, before next flight, contact Airbus to obtain approved repair instructions and accomplish those instructions accordingly.
- (3) Repair of an aeroplane as required by paragraph (2) of this AD does not constitute terminating action for the repetitive SDI as required by paragraph (1) of this AD for that aeroplane, unless specified otherwise in the instructions provided by Airbus.

**Ref. Publications:**

Airbus SB A320-53-1315 original issue dated 13 January 2016.

Airbus SB A320-53-1316 original issue dated 13 January 2016.

The use of later approved revisions of these documents is acceptable for compliance with the requirements of this AD.

**Remarks:**

1. If requested and appropriately substantiated, EASA can approve Alternative Methods of Compliance for this AD.



2. This AD was posted on 05 April 2016 as PAD 16-049 for consultation until 19 April 2016. The Comment Response Document can be found at <http://ad.easa.europa.eu>.
3. Enquiries regarding this AD should be referred to the EASA Safety Information Section, Certification Directorate. E-mail: [ADs@easa.europa.eu](mailto:ADs@easa.europa.eu).
4. For any question concerning the technical content of the requirements in this AD, please contact: AIRBUS – Airworthiness Office – EIAS; Fax +33 5 61 93 44 51;  
E-mail: [account.airworth-eas@airbus.com](mailto:account.airworth-eas@airbus.com).

